Amendments to the Claims:

Please amend the claims as shown. Applicants reserve the right to pursue any cancelled claims at a later date.

1 – 9 (cancelled)

10. (new) A turbomachine, comprising:

an axially displaceable rotor;

an annular duct between an outer guide surface fastened to an external wall and an inner guide surface arranged on the rotor;

an annular flow duct narrowing in an axial direction and formed by a working medium flowing through the annular duct;

a guide-blade ring formed from a guide blade having a guide profile extending between a platform of the guide blade arranged in the annular duct and an end of the guide blade exposed into the working medium;

a moving-blade ring formed from a moving blade having a moving profile extending between a platform of the moving blade fastened to the rotor and an end of the moving blade exposed into the working medium;

a first radial gap located in a first axial section formed between the outer guide surface and the exposed end of the moving blade; and

a second radial gap located in a second axial section which is opposite to the first axial section formed between the inner guide surface and the exposed end of the guide blade,

wherein the first and second radial gaps are parallel to a rotation axis of the rotor and a size of the radial gaps is constant over an axial displacement distance of the rotor.

11. (new) The turbomachine as claimed in claim 10, wherein the outer guide surface is formed partly by a top side of the platform of the guide blade, the top side:

facing the guide profile, and

inclined in the axial direction so that the flow duct narrows in the axial direction.

12. (new) The turbomachine as claimed in claim 10, wherein the inner guide surface is formed partly by a top side of the platform of the moving blades, the top side:

facing the moving profile, and

inclined in the axial direction so that the flow duct narrows in the axial direction.

13. (new) The turbomachine as claimed in claim 10,

wherein in the first axial section the outer guide surface is cylindrical and the inner guide surface is conically inclined relative to the rotation axis,

wherein in the second axial section the inner guide surface is cylindrical and the outer guide surface is conically inclined relative to the rotation axis, and

wherein the first and second axial sections are arranged alternatively in the axial direction.

14. (new) The turbomachine as claimed in claim 10, wherein a guide ring is configured by an axial section of the outer guide surface and is parallel to the rotation axis of the rotor.

15. (new) The turbomachine as claimed in claim 14, wherein the axial section of the outer guide surface is a sum of an axial length of the exposed end of the moving blade and the axial displacement distance of the rotor.

16. (new) The turbomachine as claimed in claim 10, wherein the turbomachine is an axial-flow compressor of a gas turbine.

17. (new) A method for improving a flow lose during an axial displacement of a rotor of a turbomachine, comprising:

arranging an annular duct between an outer guide surface fastened to an external wall and an inner guide surface arranged on the rotor;

providing a first radial gap located in a first axial section formed between the outer guide surface and an end of a moving blade fastened to the rotor and exposed into a working medium, the first radial gap parallel to a rotation axis of the rotor;

providing a second radial gap located in a second axial section which is opposite to the first axial section formed between the inner guide surface and an end of a guide blade arranged in the annular duct and exposed into the working medium, the second radial gap parallel to the rotation axis of the rotor; and

maintaining a constant size of the first and second radial gaps over a distance of the axial displacement of the rotor.

18. (new) The method as claimed in claim 17, wherein the outer guide surface is formed partly by a top side of a platform of the guide blade and the top side is inclined in the axial direction so that the annular duct narrows in the axial direction.

19. (new) The method as claimed in claim 17, wherein the inner guide surface is formed partly by a top side of a platform of the moving blade and the top side is inclined in the axial direction so that the annular duct narrows in the axial direction.

20. (new) The method as claimed in claim 17,

wherein in the first axial section the outer guide surface is cylindrical and the inner guide surface is conically inclined relative to the rotation axis,

wherein in the second axial section the inner guide surface is cylindrical and the outer guide surface is conically inclined relative to the rotation axis, and

wherein the first and second axial sections are arranged alternatively in the axial direction.

21. (new) The method as claimed in claim 17, wherein a guide ring is configured by an axial section of the outer guide surface and is parallel to the rotation axis of the rotor.

22. (new) The method as claimed in claim 21, wherein the axial section of the outer guide surface is a sum of an axial length of the exposed end of the moving blade and the axial displacement distance of the rotor.

23. (new) The method as claimed in claim 17, wherein the turbomachine is an axial-flow compressor of a gas turbine.